



Airworthiness Directive

AD No.: 2021-0227

Issued: 11 October 2021

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A318, A319, A320 and A321 aeroplanes

Effective Date: 25 October 2021

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: None

ATA 53 – Fuselage – Double Joggle Area Frame 16 and Frame 20 – Inspection / Modification

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-211, A320-212, A320-214, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231, A321-232 aeroplanes, all manufacturer serial numbers (MSN) except aeroplanes in any of the configurations below:

- Aeroplanes on which Airbus modification (mod) 160917 has been embodied in production.
- A318 aeroplanes on which Airbus mod 39195 has been embodied in production, or Airbus Service Bulletin (SB) A320-00-1219 has been embodied in service.
- A319 aeroplanes on which Airbus mod 28238, mod 28162 and mod 28342 have been embodied in production.
- Aeroplanes on which Airbus SB A320-53-1381, Airbus SB A320-53-1380, Airbus SB A320-53-1379 and Airbus SB A320-53-1378 have been embodied in service.



Definitions:

For the purpose of this AD, the following definitions apply:

Affected areas: Double joggle areas at frame (FR)16 and FR20, right-hand (RH) and left-hand (LH) sides.

The inspection SB: Airbus SB A320-53-1373 (FR16) or SB A320-53-1374 (FR20), as applicable to the affected area.

The modification SB: Airbus SB A320-53-1378 (FR20 RH), SB A320-53-1379 (FR20 LH), SB A320-53-1380 (FR16 RH) or SB A320-53-1381 (FR16 LH), as applicable to the affected area.

Reason:

During inspections accomplished in accordance with Airworthiness Limitation Item (ALI) tasks 531153 and 531155, cracks were detected in the affected areas.

This condition, if not detected and corrected, could reduce the structural integrity of the fuselage.

To address this potential unsafe condition, Airbus issued the inspection SB, to provide instructions for special detail inspection (SDI) of the affected areas, superseding the applicable ALI tasks. Airbus also issued the modification SB to provide a solution to reinforce the affected areas.

For the reason described above, this AD requires repetitive SDI of the affected areas and, depending on findings, accomplishment of applicable corrective action(s). This AD also includes reference to an optional terminating action for the repetitive SDI.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) Within the threshold as defined in Table 1 of this AD, and, thereafter, at intervals not exceeding the value as defined in Table 1 of this AD, as applicable, accomplish an SDI of each affected area in accordance with the instructions of the inspection SB.



Table 1 – SDI Compliance Time

Affected Area	Threshold (A or B, whichever occurs later)		Interval
	FR 16	A	
B		Within 6 800 FC since last accomplishment of ALI task 531153-02-1 or task 531153-02-2	
FR 20	A	Before exceeding 31 891 FC since aeroplane first flight	
	B	Within 8 900 FC since last accomplishment of ALI task 531155-02-1 or task 531155-02-2	

Corrective Action(s):

- (2) If, during any SDI as required by paragraph (1) of this AD, any crack is detected, before next flight, contact Airbus for approved repair instructions and, within the compliance time specified therein, accomplish those instructions accordingly.

Credit:

- (3) For an aeroplane that has been inspected per ALI task 531153 or 531155, and repaired using Airbus approved instructions, accomplish (repetitive) inspections of each repaired fastener hole in accordance with, and within the time period(s) after repair, as applicable, as specified in those Airbus approved repair instructions.
- (4) Any post-repair inspection instructions approved by Airbus, stating that it supersedes ALI task 531153 or task 531155 inspection requirements, supersedes the inspection requirements of paragraph (1) of this AD for the specific area detailed in the approved repair instructions statement. For all other affected areas, the requirements of this AD remain applicable.

Note 1: For all non-repaired fastener holes, see paragraph (1) or (6) of this AD, as applicable.

ALI Tasks Superseded:

- (5) Accomplishment of inspections on an aeroplane, as required by paragraphs (1) of this AD, supersedes ALI tasks 531153-02-1, 531153-02-2, 531155-02-1 and 531155-02-2, as applicable, for that aeroplane.

Acceptable Alternative Method:

- (6) Inspections on an aeroplane, accomplished per ALI tasks 531153-02-1, 531153-02-2, 531155-02-1 and 531155-02-2 during a period of up to 6 months after the effective date of this AD, are acceptable to comply with the initial requirements of paragraph (1) of this AD for that aeroplane.

Terminating Action:

- (7) Accomplishment of corrective action(s) on an aeroplane as required by paragraph (2) of this AD does not constitute terminating action for the repetitive SDI as required by paragraph (1) of this AD for that aeroplane, unless specified otherwise in the instructions provided by Airbus.



- (8) Modification of an affected area of an aeroplane in accordance with the instructions of the modification SB constitutes terminating action for the repetitive SDI as required by paragraph (1) of this AD for that affected area of that aeroplane.
- (9) Accomplishment of inspection(s) on an aeroplane, as specified by paragraph (3) of this AD for a repaired fastener hole, constitutes terminating action for the repetitive inspections as required by paragraph (1) of this AD for that repaired fastener hole.

Ref. Publications:

Airbus SB A320-53-1373 original issue dated 14 June 2018.

Airbus SB A320-53-1374 original issue dated 14 June 2018.

Airbus SB A320-53-1378 original issue dated 14 June 2018, or Revision 01 dated 17 September 2019.

Airbus SB A320-53-1379 original issue dated 14 June 2018, or Revision 01 dated 17 September 2019, or Revision 02 dated 13 January 2021.

Airbus SB A320-53-1380 original issue dated 14 June 2018, or Revision 01 dated 13 September 2019.

Airbus SB A320-53-1381 original issue dated 14 June 2018, or Revision 01 dated 18 September 2019.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 04 August 2021 as PAD 21-113 for consultation until 01 September 2021. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.



5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – IIASA; E-mail: account.airworth-eas@airbus.com.

